

Airworthiness Directive 93-24-03

Federal Register Information

Header Information

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

Docket No. 93-CE-22-AD; Amendment 39-8752; AD 93-24-03

Airworthiness Directives; Beech Models 35-33, 35-A33, 35-B33, 35-C33, E33, F33, G33, 35-C33A, E33A, F33A, E33C, F33C, 36, A36, A36TC and B36TC Airplanes

Preamble Information

AGENCY: Federal Aviation Administration, DOT

DATES: Effective January 21, 1994

Regulatory Information

93-24-03 BEECH AIRCRAFT CORPORATION: Amendment 39-8752; Docket No. 93-CE- 22-AD. Supersedes AD 92-15-06, Amendment 39-8300 which superseded AD 91-23-07, Amendment 39-8076.

Applicability: The following Beech model and serial numbered airplanes, certificated in any category:

MODELS	SERIAL NUMBERS
35-33, 35-A33, 35-B33, 35-C33, E33, F33, and G33	CD-1 through CD-1304
35-C33A, E33A, and F33A	CE-1 through CE-1425
E33C and F33C	CJ-1 through CJ-179
36 and A36	E-1 through E-2518
A36TC and B36TC	EA-1 through EA-500

Compliance: Required as indicated after the effective date of this AD, unless already accomplished (compliance with superseded AD 92-15-06 or superseded AD 91-23-07).

To prevent separation of the rudder from the airplane caused by cracks in the forward rudder spar, accomplish the following:

(a) Upon the accumulation of 1,000 hours time-in-service (TIS) or within the next 100 hours TIS, whichever occurs later, inspect the rudder forward spar for cracks in accordance with the instructions in Beech Service Bulletin (SB) No. 2333, Revision 1, dated November 1991.

(b) If no cracks are found, accomplish one of the following:

(1) Reinspect the rudder forward spar for cracks in accordance with the instructions in Beech SB No. 2333, Revision 1, dated November 1991, at intervals not to exceed 500 hours TIS until either paragraph (b)(2), (b)(3), or (b)(4) of this AD is accomplished;

(2) Install Kit No. 33-6001-1 S in accordance with Beech SB No. 2333, Revision 1, dated November 1991;

(3) Install a Spacecraft Machine Products (SMP) rudder spar upper-hinge reinforcement bracket in accordance with Supplemental Type Certificate (STC) SA4899NM; or

(4) Replace the rudder assembly with either part number 33-630000-137, - 139, -141, -167, or -169, as applicable, in accordance with the instructions in Beech SB No. 2333, Revision 1, dated November 1991.

(c) If cracks are found, prior to further flight, accomplish one of the following:

(1) Replace the rudder assembly with either part number 33-630000-137, - 139, -141, -167, or -169, as applicable, in accordance with the instructions in Beech SB No. 2333, Revision 1, dated November 1991;

(2) Install Kit No. 33-6001-1 S in accordance with Beech SB No. 2333, Revision 1, dated November 1991; or

(3) If the cracks are found in the area of the upper hinge, the middle hinge, or both the upper and middle hinge as specified in Beech SB No. 2333, Revision 1, dated November 1991, then stop drill the cracks and install an SMP upper-hinge reinforcement bracket in accordance with STC SA4899NM. For cracks in the middle hinge, install the upper-hinge reinforcement bracket and also install an SMP rudder spar middle-hinge reinforcement bracket in accordance with STC SA5870NM.

(d) If a modification or replacement has been accomplished in accordance with either paragraph (b)(2), (b)(3), (b)(4),

(c)(1), (c)(2), or (c)(3) of this AD, then no repetitive inspections are required by this AD.

(e) Special flight permits may be issued in accordance with FAR 21.197 and 21.199 to operate the airplane to a location where the requirements of this AD can be accomplished.

(f) An alternative method of compliance or adjustment of the initial or repetitive compliance times that provides an equivalent level of safety may be approved by the Manager, Wichita Aircraft Certification Office, FAA, 1801 Airport Road, Mid-Continent Airport, Wichita, Kansas 67209. The request shall be forwarded through an appropriate FAA Maintenance Inspector, who may add comments and then send it to the Manager, Wichita Aircraft Certification Office.

NOTE: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Wichita Aircraft Certification Office.

(g) The inspections, installations, or replacements required by this AD shall be done in accordance with Beech Service Bulletin No. 2333, Revision 1, dated November 1991. This incorporation by reference was previously approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR Part 51 on August 22, 1992. Copies may be obtained from Beech Aircraft Corporation, P.O. Box 85, Wichita, Kansas 67201-0085. Copies may be inspected at the FAA, Central Region, Office of the Assistant Chief Counsel, Room 1558, 601 E. 12th Street, Kansas City, Missouri, or at

the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(h) This amendment (39-8752) supersedes AD 92-15-06, Amendment 39-8300 which superseded AD 91-23-07, Amendment 39-8076.

(i) This amendment (39-8752) becomes effective on January 21, 1994.

Footer Information

Comments

Completed 11 March 1992 with superseded AD 91-23-07 (Beech SB No. 2333, Revision 1)